

a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) if promulgated, will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A copy of the draft regulatory evaluation prepared for this action is contained in the Rules Docket. A copy of it may be obtained by contacting the Rules Docket at the location provided under the caption **ADDRESSES**.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Safety.

The Proposed Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration proposes to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 USC 106(g), 40101, 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by adding the following new airworthiness directive:

Boeing: Docket 95–NM–98–AD.

Applicability: Model 747–400 series airplanes powered by General Electric CF6–80C2 or Pratt & Whitney PW4000 series engines; as identified in Boeing Alert Service Bulletin 747–28A2185, Revision 1, dated September 21, 1995, and Boeing Service Bulletin 747–28–2146, dated August 13, 1992; certificated in any category.

Note 1: This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been otherwise modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must use the authority provided in paragraph (b) of this AD to request approval from the FAA. This approval may address either no action, if the current configuration eliminates the unsafe condition; or different actions necessary to address the unsafe condition described in this AD. Such a request should include an assessment of the effect of the changed configuration on the unsafe condition addressed by this AD. In no case does the presence of any modification, alteration, or repair remove any airplane from the applicability of this AD.

Compliance: Required as indicated, unless accomplished previously.

To prevent fracturing of the coupling nut, which could result in release of fuel onto the engine cowling and a subsequent fire, accomplish the following:

(a) Within 18 months after the effective date of this AD, accomplish the requirements of paragraph (a)(1) or (a)(2), as applicable.

(1) For Model 747–400 series airplanes identified in Boeing Alert Service Bulletin 747–28A2185, Revision 1, dated September 21, 1995: Replace the strut fuel tubes and couplings at engine numbers 1 and 4 with new redesigned (shrouded) couplings, in accordance with that alert service bulletin.

(2) For Model 747–400 series airplanes having variable numbers RT641 through RT650 inclusive, identified in Boeing Service Bulletin 747–28–2146, dated August 13, 1992: On engine positions 1 and 4 only, install new fuel lines, shrouded fuel line couplings (between the strut mid bulkhead and the wing front spar), and drain lines in accordance with that service bulletin.

(b) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Seattle Aircraft Certification Office (ACO), FAA, Transport Airplane Directorate. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, Seattle ACO.

Note 2: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Seattle ACO.

(c) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

Issued in Renton, Washington, on December 6, 1995.

Darrell M. Pederson,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 95–30213 Filed 12–11–95; 8:45 am]

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14 CFR Part 39

[Docket No. 95–NM–78–AD]

Airworthiness Directives; Airbus Model A300–600 Series Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: This document proposes the adoption of a new airworthiness directive (AD) that is applicable to certain Airbus Model A300–600 series airplanes. This proposal would require inspections to detect corrosion and cracking of the lower horizontal-stabilizer cutout longeron, the corner fitting, the skin strap, and the outer skin; and repair, if necessary. This

proposal is prompted by cracking found at the lower corner of the horizontal-stabilizer cutout longeron during a full scale fatigue test. The actions specified by the proposed AD are intended to prevent such cracking, which could result in reduced structural integrity of the horizontal stabilizer cutout longeron.

DATES: Comments must be received by January 23, 1996.

ADDRESSES: Submit comments in triplicate to the Federal Aviation Administration (FAA), Transport Airplane Directorate, ANM–103, Attention: Rules Docket No. 95–NM–78–AD, 1601 Lind Avenue, SW., Renton, Washington 98055–4056. Comments may be inspected at this location between 9:00 a.m. and 3:00 p.m., Monday through Friday, except Federal holidays.

The service information referenced in the proposed rule may be obtained from Airbus Industrie, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France. This information may be examined at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington.

FOR FURTHER INFORMATION CONTACT: Charles Huber, Aerospace Engineer, Standardization Branch, ANM–113, FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington 98055–4056; telephone (206) 277–2589; fax (206) 227–1149.

SUPPLEMENTARY INFORMATION:

Comments Invited

Interested persons are invited to participate in the making of the proposed rule by submitting such written data, views, or arguments as they may desire. Communications shall identify the Rules Docket number and be submitted in triplicate to the address specified. All communications received on or before the closing date for comments, specified above, will be considered before taking action on the proposed rule. The proposals contained in this notice may be changed in light of the comments received.

Comments are specifically invited on the overall regulatory, economic, environmental, and energy aspects of the proposed rule. All comments submitted will be available, both before and after the closing date for comments, in the Rules Docket for examination by interested persons. A report summarizing each FAA-public contact concerned with the substance of this proposal will be filed in the Rules Docket.

Commenters wishing the FAA to acknowledge receipt of their comments

submitted in response to this notice must submit a self-addressed, stamped postcard on which the following statement is made: "Comments to Docket Number 95-NM-78-AD." The postcard will be date stamped and returned to the commenter.

Availability of NPRMs

Any person may obtain a copy of this NPRM by submitting a request to the FAA Transport Airplane Directorate, ANM-103, Attention: Rules Docket No. 95-NM-78-AD, 1601 Lind Avenue, SW., Renton, Washington 98055-4056.

Discussion

The Direction Générale de l'Aviation Civile (DGAC), which is the airworthiness authority for France, recently notified the FAA that an unsafe condition may exist on certain Airbus Model A300-600 series airplanes. The DGAC advises that, during a full-scale fatigue test, cracking was found on the test airplane at the lower corner of the horizontal-stabilizer cutout longeron, between frame (FR) 87 and FR89 and between stringer (STGR) 24 and STGR27, left- and right-hand. The cracking was found after 87,675 simulated flights. Such fatigue cracking, if not corrected, could result in reduced structural integrity of the horizontal stabilizer cutout longeron.

Airbus has issued Service Bulletin A300-53-6042, Revision 1, dated February 20, 1995, which describes procedures for repetitive visual and eddy current inspections to detect corrosion and cracking of the lower horizontal-stabilizer cutout longeron, the corner fitting, the skin strap, and the outer skin between FR87 and FR89 and between STGR24 and STGR27, left- and right-hand; and repair, if necessary. The service bulletin also describes procedures for repetitive rotating probe inspections to detect cracks in the fastener holes at the same locations, and repair, if necessary. The service bulletin also describes procedures for certain follow-on actions, which include installing a new corner fitting, installing a new longeron, and performing a cold working procedure repairing cracks of certain measurements. The DGAC classified this service bulletin as mandatory and issued French airworthiness directive (CN) 94-269-171(B)R1, dated March 29, 1995, in order to assure the continued airworthiness of these airplanes in France.

This airplane model is manufactured in France and is type certificated for operation in the United States under the provisions of section 21.29 of the Federal Aviation Regulations (14 CFR

21.29) and the applicable bilateral airworthiness agreement. Pursuant to this bilateral airworthiness agreement, the DGAC has kept the FAA informed of the situation described above. The FAA has examined the findings of the DGAC, reviewed all available information, and determined that AD action is necessary for products of this type design that are certificated for operation in the United States.

Since an unsafe condition has been identified that is likely to exist or develop on other airplanes of the same type design registered in the United States, the proposed AD would require repetitive visual and eddy current inspections to detect corrosion and fatigue cracking of the lower horizontal-stabilizer cutout longeron, the corner fitting, the skin strap, and the skin between FR87 and FR89 and between STGR24 and STGR27, left-hand and right-hand. The proposed AD would also require repetitive rotating probe inspections to detect cracks in the fastener holes at the same locations; and repair or certain a follow-on actions, if necessary. The actions would be required to be accomplished in accordance with the service bulletin described previously. Certain repairs would be required to be accomplished in accordance with a method approved by the FAA.

Operators should note that certain requirements of this proposed AD would differ from actions recommended in the referenced Airbus service bulletin. The service bulletin specifies that inspection thresholds and intervals may be adjusted based on certain average flight operations of the airplane. However, the FAA has determined that such adjustments would not address the unsafe condition in a timely manner. Therefore, this proposed AD does not permit such adjustments. In developing the appropriate compliance time for the proposed rule, the FAA considered not only the manufacturer's recommendation, but the safety implications involved with cracking of the horizontal-stabilizer cutout longeron and the number of landings that had been accumulated when cracking was detected. In light of these factors, the FAA finds the compliance times specified in the proposed AD for initiating the required actions to be warranted, in that they represent an appropriate interval of time allowable for the affected airplanes to continue to operate without compromising safety.

Additionally, the service bulletin specifies that operators need not count touch-and-go landings in determining the total number of landings between two consecutive inspections, even if

those landings are less than five percent of the landings between inspection intervals. Since the fatigue cracking that was found in the lower horizontal stabilizer cutout longeron is aggravated by landing, the FAA finds that all touch-and-go landings must be counted in determining the total number of landings between two consecutive inspections.

The FAA estimates that 2 airplanes of U.S. registry would be affected by this proposed AD, that it would take approximately 268 work hours per airplane to accomplish the proposed actions, and that the average labor rate is \$60 per work hour. Based on these figures, the cost impact of the proposed AD on U.S. operators is estimated to be \$32,160 to \$16,080 per airplane.

The cost impact figure discussed above is based on assumptions that no operator has yet accomplished any of the proposed requirements of this AD action, and that no operator would accomplish those actions in the future if this AD were not adopted.

The regulations proposed herein would not have substantial direct effects on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, in accordance with Executive Order 12612, it is determined that this proposal would not have sufficient federalism implications to warrant the preparation of a Federalism Assessment.

For the reasons discussed above, I certify that this proposed regulation (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) if promulgated, will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A copy of the draft regulatory evaluation prepared for this action is contained in the Rules Docket. A copy of it may be obtained by contacting the Rules Docket at the location provided under the caption **ADDRESSES**.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Safety.

The Proposed Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration proposes to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. App. 1354(a), 1421 and 1423; 49 U.S.C. 106(g); and 14 CFR 11.89.

§ 39.13 [Amended]

2. Section 39.13 is amended by adding the following new airworthiness directive:

Airbus Industries: Docket 95–NM–78–AD.

Applicability: Model A300–600 series airplanes on which Airbus Modification No. 6146 has not been installed, certificated in any category.

Note 1: This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must use the authority provided in paragraph (f) of this AD to request approval from the FAA. This approval may address either no action, if the current configuration eliminates the unsafe condition; or different actions necessary to address the unsafe condition described in this AD. Such a request should include an assessment of the effect of the changed configuration on the unsafe condition addressed by this AD. In no case does the presence of any modification, alternation, or repair remove any airplane from the applicability of this AD.

Compliance: Required as indicated, unless accomplished previously.

To prevent reduced structural integrity of the horizontal stabilizer cutout longeron due to fatigue cracking, accomplish the following:

(a) Prior to the accumulation of 18,000 total landings, or within 2,000 landings after the effective date of this AD, whichever occurs later: Perform a visual and an eddy current inspection to detect cracks and/or corrosion of Areas 1 and 2 of the lower horizontal stabilizer cutout longeron, as defined in Airbus Service Bulletin A300–53–6042, Revision 1, and dated February 20, 1995. Perform the inspections in accordance with the service bulletin.

(b) Perform a visual and an eddy current inspection to detect cracks and/or corrosion of Area 3 of the lower horizontal-stabilizer cutout longeron, as defined in Airbus Service Bulletin A300–53–6042, Revision 1, dated February 20, 1995. Perform these inspections in accordance with the service bulletin, at the later of the times specified in paragraph (b)(1) and (b)(2) of this AD.

(1) Prior to the accumulation of 24,000 total landings, but not before the accumulation of 18,000 total landings; or

(2) Prior to the accumulation of 2,000 landings after the effective date of this AD.

(c) If no cracking is detected during any inspection required by this AD: Prior to further flight, cold work and ream the vacated fastener holes, in accordance with

Airbus Service Bulletin A300–53–6042, Revision 1, dated February 20, 1995; and perform the requirements of paragraph (c)(1) or (c)(2) of this AD, as applicable.

(1) For airplanes on which no cracking is found in Area 1 or 2: Repeat the inspections required by paragraph (a) of this AD thereafter at intervals not to exceed 6,000 flight cycles.

(2) For airplanes on which no cracking is found in Area 3: Perform the various follow-on actions in accordance with the service bulletin. (The follow-on actions include installing a new corner fitting, installing a new longeron and performing a cold working procedure.) After accomplishment of these follow-on actions, no further action is required by this AD.

(d) If any cracking is detected during any inspection required by this AD, perform the requirements of paragraph (d)(1) or (d)(2) of this AD, as applicable.

(1) If any cracking is found in Area 1 or 3 that is within the limits specified in Airbus Service bulletin A300–53–6042, Revision 1, dated February 20, 1995: Prior to further flight, repair in accordance with the service bulletin.

(2) If any cracking is found in Area 2, or if any cracking is found in any area and that cracking is beyond the limits described in Airbus Service Bulletin A300–53–6042, Revision 1, dated February 20, 1995: Prior to further flight, repair in accordance with a method approved by the Manager, Standardization Branch, ANM–113, FAA, Transport Airplane Directorate.

(e) If any corrosion is detected during any inspection required by this AD, prior to further flight, repair the corrosion in accordance with Airbus Service Bulletin A300–53–6042, Revision 1, dated February 20, 1995.

(f) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Standardization Branch, ANM–113. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, Standardization Branch, ANM–113.

Note 2: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Standardization Branch, ANM–113.

(g) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

Issued in Renton, Washington, on December 6, 1995.

Darrell M. Pederson,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 95–30212 Filed 12–11–95; 8:45 am]

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FEDERAL COMMUNICATIONS COMMISSION

47 CFR Parts 64 and 68

[CC Docket No. 87–124; FCC 95–474]

Access to Telecommunications Equipment and Services by Persons With Disabilities (Hearing Aid Compatibility)

AGENCY: Federal Communications Commission.

ACTION: Proposed rule.

SUMMARY: On November 28, 1995, the Commission adopted a Notice of Proposed Rulemaking (NPRM) regarding hearing aid compatibility of wireline telephones. Rules proposed in the NPRM would require that all wireline telephones in the workplace, confined settings (e.g., hospitals, nursing homes) and hotels and motels eventually would be hearing aid compatible and have volume control. This NPRM contains proposed or modified information collections subject to the Paperwork Reduction Act of 1995 (PRA). It has been submitted to the Office of Management and Budget (OMB) for review under Section 3507(d) of the PRA. OMB, the general public, and other Federal agencies are invited to comment on the proposed or modified information collections contained in this proceeding.

DATES: Written comments by the public on the NPRM and on the proposed and/or modified information collections are due on or before January 12, 1996, and reply comments are due on or before February 16, 1996. Written comments must be submitted by the Office of Management and Budget (OMB) on the proposed and/or modified information collections on or before February 12, 1996.

ADDRESSES: Office of the Secretary, Room 222, Federal Communications Commission, 1919 M Street, NW, Washington, DC 20554. In addition to filing comments with the Secretary, a copy of any comments on the information collections contained herein should be submitted to Dorothy Conway, Federal Communications Commission, Room 234, 1919 M Street, N.W., Washington, DC 20554, or via the Internet to dconway@fcc.gov, and to Timothy Fain, OMB Desk Officer, 10236 NEOB, 725–17th Street, N.W., Washington, DC 20503 or via the Internet to fain-t@al.eop.gov.

FOR FURTHER INFORMATION CONTACT: Greg Lipscomb, Attorney, 202/418–2340, Fax 202/418–2345, TTY 202/418–0484, glipscom@fcc.gov, Network Services