(b) As of the effective date of this AD, no person shall install on any airplane a Dunlop Main Undercarriage Ram, part number (P/N) AC67132, AC67134, AC67848, or AC67850, unless it has been modified in accordance with Fokker Service Bulletin F27/32–168, dated October 23, 1996.

(c) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, International Branch, ANM–116, FAA, Transport Airplane Directorate. Operators shall submit their request through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, International Branch, ANM–116.

Note 3: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the International Branch, ANM–116.

(d) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

(e) The replacement shall be done in accordance with Fokker Service Bulletin F27/32–168, dated October 23, 1996. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Fokker Services B.V., Technical Support Department, P.O. Box 75047, 1117 ZN Schiphol Airport, the Netherlands. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

Note 4: The subject of this AD is addressed in Dutch airworthiness directive 1996–142(A), dated November 29, 1996.

(f) This amendment becomes effective on July 20, 1998.

Issued in Renton, Washington, on June 5, 1998.

Darrell M. Pederson,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 98–15677 Filed 6–12–98; 8:45 am]

BILLING CODE 4910–13–U

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 97–NM–194–AD; Amendment 39–10586; AD 98–12–33]

RIN 2120–AA64

Airworthiness Directives; Airbus Model A320 Series Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule.

SUMMARY: This amendment adopts a new airworthiness directive (AD), applicable to certain Airbus Model A320 series airplanes, that requires repetitive inspections to detect fatigue cracking on the connecting angle between frame 56 and the right-hand frame support at stringer 38; and replacement of the connecting angle, if necessary. This amendment also provides for an optional terminating action for the repetitive inspections.

This amendment is prompted by issuance of mandatory continuing airworthiness information by a foreign civil airworthiness authority. The actions specified by this AD are intended to detect and correct fatigue cracking on the connecting angle, which could result in reduced structural integrity of the airplane.

DATES: Effective July 20, 1998.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of July 20, 1998.

ADDRESSES: The service information referenced in this AD may be obtained from Airbus Industrie, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France. This information may be examined at the Federal Aviation Administration (FAA), Transport Airplane Directorate, Rules Docket, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

FOR FURTHER INFORMATION CONTACT: Norman B. Martenson, Manager, International Branch, ANM–116, FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington 98055–4056; telephone (425) 227–2110; fax (425) 227–1149.

SUPPLEMENTARY INFORMATION: A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an airworthiness directive (AD) that is applicable to certain Airbus Model A320 series airplanes was published in the **Federal Register** on April 14, 1998 (63 FR 18156). That action proposed to require repetitive inspections to detect fatigue cracking on the connecting angle between frame 56 and the right-hand frame support at stringer 38; and replacement of the connecting angle, if necessary. That action also proposed to provide for an optional terminating action for the repetitive inspections.

Comments

Interested persons have been afforded an opportunity to participate in the making of this amendment. Due consideration has been given to the two comments received.

The commenters support the proposed rule.

Conclusion

After careful review of the available data, including the comments noted above, the FAA has determined that air safety and the public interest require the adoption of the rule as proposed.

Cost Impact

The FAA estimates that 5 airplanes of U.S. registry will be affected by this AD. It will take approximately 1 work hour per airplane to accomplish the required inspection, at an average labor rate of $60 per work hour. Based on this figure, the cost impact of the inspection required by this AD on U.S. operators is estimated to be $300, or $60 per airplane, per inspection cycle.

The cost impact figure discussed above is based on assumptions that no operator has yet accomplished any of the requirements of this AD action, and that no operator would accomplish those actions in the future if this AD were not adopted.

Should an operator elect to perform the optional terminating replacement provided by this AD, it would take approximately 3 work hours per airplane to accomplish the modification, at an average labor rate of $60 per work hour. Required parts would cost $136 or $153 per airplane, depending on the service kit purchased. Based on these figures, the cost impact of the optional terminating modification provided by this AD on U.S. operators is estimated to be as low as $1,580, or $316 per airplane, and as high as $1,665, or $333 per airplane.

Regulatory Impact

The regulations adopted herein will not have substantial direct effects on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, in accordance with Executive Order 12612, it is determined that this final rule does not have sufficient federalism implications to warrant the preparation of a Federalism Assessment.

For the reasons discussed above, I certify that this action (1) is not a ''significant regulatory action'' under Executive Order 12866; (2) is not a ''significant rule'' under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities

under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained from the Rules Docket at the location provided under the caption **ADDRESSES**.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by adding the following new airworthiness directive:

98–12–33 Airbus Industrie: Amendment 39–10586. Docket 97–NM–194–AD.

Applicability: Model A320 series airplanes, on which Airbus Modification 20941 (reference Airbus Service Bulletin A320–53–1011, dated December 9, 1994) has not been accomplished, certificated in any category.

Note 1: This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been otherwise modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (c) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance: Required as indicated, unless accomplished previously.

To detect and correct fatigue cracking on the connecting angle between frame 56 and the right-hand frame support at stringer 38, which could result in reduced structural integrity of the airplane, accomplish the following:

(a) Prior to the accumulation of 20,000 total flight cycles, or within 1,000 flight cycles after the effective date of this AD, whichever occurs later, perform a visual inspection for fatigue cracking on the connecting angle between frame 56 and the right-hand frame support at stringer 38, in accordance with Airbus Service Bulletin A320–53–1084, Revision 1, dated November 28, 1995.

(1) If no cracking is detected, accomplish either paragraph (a)(1)(i) or (a)(1)(ii) of this AD.

(i) Prior to further flight, replace the connecting angle between frame 56 and the right-hand frame support at stringer 38 with a new part, in accordance with Airbus Service Bulletin A320–53–1011, dated December 9, 1994; or

(ii) Repeat the visual inspection thereafter at intervals not to exceed 12,000 flight cycles.

(2) If any cracking is detected, prior to further flight, replace the connecting angle between frame 56 and the right-hand frame support at stringer 38 with a new part, in accordance with Airbus Service Bulletin A320–53–1011, dated December 9, 1994.

(b) Accomplishment of the replacement of the connecting angle constitutes terminating action for the repetitive inspection requirements of this AD.

(c) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, International Branch, ANM–116, FAA, Transport Airplane Directorate. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, International Branch, ANM–116.

Note 2: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the International Branch, ANM–116.

(d) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

(e) The actions shall be done in accordance with the following Airbus service bulletins, which contain the specified effective pages:

Service bulletin referenced and date	Page No. shown on page	Revision level shown on page	Date shown on page
A320–53–1084, Revision 1, November 28, 1995	1, 2	1	Nov. 28, 1995.
	3–11	Original	Dec. 9, 1994.
A320–53–1011, December 9, 1994	1–11	Original	Dec. 9, 1994.

This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Airbus Industrie, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

Note 3: The subject of this AD is addressed in French airworthiness directive 96–237–090(B), dated October 23, 1996, and Erratum to French airworthiness directive 96–237–090(B), dated February 26, 1997.

(f) This amendment becomes effective on July 20, 1998.

Issued in Renton, Washington, on June 5, 1998.

Darrell M. Pederson,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 98–15679 Filed 6–12–98; 8:45 am]

BILLING CODE 4910–13–U

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 97–NM–64–AD; Amendment 39–10589; AD 98–13–01]

RIN 2120–AA64

Airworthiness Directives; Aerospatiale Model ATR42 and ATR72 Series Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule.

SUMMARY: This amendment adopts a new airworthiness directive (AD), applicable to certain Aerospatiale Model ATR42 and ATR72 series airplanes, that