

under the criteria of the Regulatory Flexibility Act.

You can find our regulatory evaluation and the estimated costs of compliance in the AD Docket.

#### List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Safety.

#### The Proposed Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA proposes to amend 14 CFR part 39 as follows:

#### PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

#### § 39.13 [Amended]

2. The FAA amends § 39.13 by adding the following new AD:

**Saab AB, Saab Aerosystems:** Docket No. FAA–2010–0042; Directorate Identifier 2009–NM–010–AD.

#### Comments Due Date

(a) We must receive comments by March 7, 2011.

#### Affected ADs

(b) None.

#### Applicability

(c) This AD applies to the Saab AB, Saab Aerosystems airplanes, certificated in any category, identified in paragraphs (c)(1) and (c)(2) of this AD, that have been modified in accordance with Supplemental Type Certificate (STC) ST00224WI–D, ST00146WI–D, or SA984GL–D.

(1) Model SAAB 340A (SAAB/SF340A) airplanes, serial numbers 004 through 159 inclusive.

(2) Model SAAB 340B airplanes, serial numbers 160 through 459 inclusive.

#### Subject

(d) Joint Aircraft System Component (JASC)/Air Transport Association (ATA) of America Code 53: Fuselage.

#### Unsafe Condition

(e) This AD results from a report of a crack found behind the external adapter plate of the antennae during inspection. Similar cracking was found on two additional airplanes, and extensive corrosion was found on one airplane. The Federal Aviation Administration is issuing this AD to detect and correct corrosion and cracking behind the external adapter plate of the antennae of certain damage-tolerant structure, which could result in reduced structural integrity and consequent rapid depressurization of the airplane.

#### Compliance

(f) You are responsible for having the actions required by this AD performed within the compliance times specified.

#### Inspection/Corrective Actions

(g) Within 600 flight cycles after the effective date of this AD: Remove the external adapter plate of the antennae installation and do a general visual inspection of the fuselage surface for corrosion and cracking behind the external adapter plate of the antennae installation. If any corrosion or cracking is found, repair before further flight. If no corrosion or cracking is found, before further flight, ensure that proper corrosion protection has been applied before reinstalling the adapter plate. Do all the actions required by this paragraph in accordance with a method approved by the Manager, Wichita Aircraft Certification Office (ACO), FAA.

**Note 1:** For the purposes of this AD, a general visual inspection is: “A visual examination of an interior or exterior area, installation, or assembly to detect obvious damage, failure, or irregularity. This level of inspection is made from within touching distance unless otherwise specified. A mirror may be necessary to ensure visual access to all surfaces in the inspection area. This level of inspection is made under normally available lighting conditions such as daylight, hangar lighting, flashlight, or droplight and may require removal or opening of access panels or doors. Stands, ladders, or platforms may be required to gain proximity to the area being checked.”

#### Reporting Requirement

(h) At the applicable time specified in paragraph (h)(1) or (h)(2) of this AD: Submit a report of the positive findings of the inspections required by paragraph (g) of this AD. Send the report to the Manager, Wichita ACO. The report must contain, at a minimum, the inspection results, a description of any discrepancies found, the airplane serial number, and the number of flight cycles and flight hours on the airplane since installation of the STC. Under the provisions of the Paperwork Reduction Act (44 U.S.C. 3501 *et seq.*), the Office of Management and Budget (OMB) has approved the information collection requirements contained in this AD and has assigned OMB Control Number 2120–0056.

(1) If the inspection was done on or after the effective date of this AD: Submit the report within 30 days after the inspection.

(2) If the inspection was done before the effective date of this AD: Submit the report within 30 days after the effective date of this AD.

(3) A Federal agency may not conduct or sponsor, and a person is not required to respond to, nor shall a person be subject to a penalty for failure to comply with a collection of information subject to the requirements of the Paperwork Reduction Act unless that collection of information displays a current valid OMB Control Number. The OMB Control Number for this information collection is 2120–0056. Public reporting for this collection of information is estimated to be approximately 5 minutes per response, including the time for reviewing instructions, completing and reviewing the collection of information. All responses to this collection of information are mandatory. Comments concerning the accuracy of this

burden and suggestions for reducing the burden should be directed to the FAA at: 800 Independence Ave., SW, Washington, DC 20591, Attn: Information Collection Clearance Officer, AES–200.

#### Special Flight Permit

(i) Special flight permits, as described in Section 21.197 and Section 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199), may be issued to operate the airplane to a location where the requirements of this AD can be accomplished, but concurrence by the Manager, Wichita ACO, FAA, is required prior to issuance of the special flight permit.

#### Alternative Methods of Compliance (AMOCs)

(j)(1) The Manager, Wichita ACO, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. Send information to ATTN: William Griffith, Aerospace Engineer, Airframe Branch, ACE–118W, FAA, Wichita ACO, 1801 Airport Road, Room 100, Mid-Continent Airport, Wichita, Kansas 67209; telephone (316) 946–4116; fax (316) 946–4107.

(2) To request a different method of compliance or a different compliance time for this AD, follow the procedures in 14 CFR 39.19. Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office. The AMOC approval letter must specifically reference this AD.

Issued in Renton, Washington on January 12, 2011.

**Jeffrey E. Duven,**

*Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.*

[FR Doc. 2011–1118 Filed 1–19–11; 8:45 am]

**BILLING CODE 4910–13–P**

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. FAA–2011–0026; Directorate Identifier 2010–NM–104–AD]

RIN 2120–AA64

#### Airworthiness Directives; The Boeing Company Model 777–200 and –300 Series Airplanes Equipped With Pratt and Whitney Engines

**AGENCY:** Federal Aviation Administration (FAA), DOT.

**ACTION:** Notice of proposed rulemaking (NPRM).

**SUMMARY:** We propose to adopt a new airworthiness directive (AD) for certain Model 777–200 and –300 series airplanes. This proposed AD would require repetitive inspections for

hydraulic fluid contamination of the interior of the strut disconnect assembly; repetitive inspections for discrepancies of the interior of the strut disconnect assembly, if necessary; repetitive inspections of the exterior of the strut disconnect assembly for cracks, if necessary; and corrective action if necessary. This proposed AD also provides an optional terminating action for the inspections. This proposed AD results from reports of system disconnect boxes that have been contaminated with hydraulic fluid and, in one incident, led to subsequent cracking of titanium parts in the system disconnect assembly. We are proposing this AD to detect and correct hydraulic fluid contamination, which can cause cracking of titanium parts in the system disconnect assembly, resulting in compromise of the engine firewall. A cracked firewall can allow fire in the engine area to enter the strut and can lead to an uncontained engine strut fire if flammable fluid is present. Cracking of the disconnect box may also reduce the effectiveness of the fire extinguishing system in the engine compartment and could contribute to an uncontained engine fire. In addition, a cracked disconnect box can leak flammable fluids into the engine core, which can initiate an engine fire, and lead to one or both fire conditions discussed above.

**DATES:** We must receive comments on this proposed AD by March 7, 2011.

**ADDRESSES:** You may send comments by any of the following methods:

- **Federal eRulemaking Portal:** Go to <http://www.regulations.gov>. Follow the instructions for submitting comments.
- **Fax:** 202-493-2251.
- **Mail:** U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue, SE., Washington, DC 20590.
- **Hand Delivery:** U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue, SE., Washington, DC 20590, between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

For service information identified in this proposed AD, contact Boeing Commercial Airplanes, Attention: Data & Services Management, P.O. Box 3707, MC 2H-65, Seattle, Washington 98124-2207; telephone 206-544-5000, extension 1; fax 206-766-5680; e-mail [me.boecom@boeing.com](mailto:me.boecom@boeing.com); Internet <https://www.myboeingfleet.com>. You may review copies of the referenced service information at the FAA, Transport Airplane Directorate, 1601

Lind Avenue, SW., Renton, Washington. For information on the availability of this material at the FAA, call 425-227-1221.

#### Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov>; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this proposed AD, the regulatory evaluation, any comments received, and other information. The street address for the Docket Office (telephone 800-647-5527) is in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

#### FOR FURTHER INFORMATION CONTACT:

Kevin Nguyen, Propulsion Branch, ANM-140S, FAA, Seattle Aircraft Certification Office (ACO), 1601 Lind Avenue, SW., Renton, Washington 98057-3356; telephone (425) 917-6501; fax (425) 917-6590.

#### SUPPLEMENTARY INFORMATION:

##### Comments Invited

We invite you to send any written relevant data, views, or arguments about this proposed AD. Send your comments to an address listed under the **ADDRESSES** section. Include "Docket No. FAA-2011-0026; Directorate Identifier 2010-NM-104-AD" at the beginning of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of this proposed AD. We will consider all comments received by the closing date and may amend this proposed AD because of those comments.

We will post all comments we receive, without change, to <http://www.regulations.gov>, including any personal information you provide. We will also post a report summarizing each substantive verbal contact we receive about this proposed AD.

##### Discussion

We have received reports of system disconnect boxes that have been contaminated with hydraulic fluid. One operator has found cracks in the system disconnect assembly bulkheads and lower skin panel. Subsequent analysis at Boeing found hydrogen embrittlement of the system disconnect assembly bulkheads and lower skin panel. The system disconnect assembly is made from titanium and is located near the hot engine core of the engine where temperatures can exceed 270 degrees Fahrenheit. The presence of hydraulic fluid and temperature above 270 degrees

Fahrenheit can result in hydrogen embrittlement of the titanium system disconnect assembly. Hydrogen embrittlement combined with a high sonic vibration environment can result in cracking of the system disconnect assembly. The system disconnect assembly is a box where hydraulic, fuel, and electrical connections are made between the engine and the strut. This box acts as a firewall between the engine compartment and the strut. The engine compartment has a fire extinguishing system while the strut does not. The strut is considered a flammable leakage zone where flammable fluids may be present. These fluids are subsequently drained from the strut and system disconnect box. A cracked firewall can allow fire in the engine area to enter the strut and can lead to an uncontained engine strut fire if flammable fluid is present. Cracking of the disconnect box may also reduce the effectiveness of the fire extinguishing system in the engine compartment and could contribute to an uncontained engine fire. In addition, a cracked disconnect box can leak flammable fluids into the engine core, which can initiate an engine fire, and lead to one or both fire conditions discussed above.

##### Relevant Service Information

We have reviewed Boeing Alert Service Bulletin 777-54A0024, Revision 1, dated November 4, 2010. This service bulletin describes procedures for doing repetitive general visual inspections for hydraulic fluid contamination of the interior of the strut disconnect assembly; repetitive detailed inspections for discrepancies (e.g., hydraulic fluid coking, heat discoloration, cracks, and etching or pitting) of the interior of the strut disconnect assembly, if certain conditions are found; repetitive detailed inspections of the exterior of the strut disconnect assembly for cracks; and corrective action, if certain conditions are found. The corrective action is replacing the titanium system disconnect assembly with an Inconel system disconnect assembly. If accomplished, the replacement will eliminate the potential for hydrogen embrittlement and subsequent cracking, and would eliminate the need for the inspections of the titanium strut disconnect assembly.

##### FAA's Determination and Requirements of This Proposed AD

We are proposing this AD because we evaluated all relevant information and determined the unsafe condition described previously is likely to exist or

develop in other products of these same type designs. This proposed AD would require accomplishing the actions specified in the service information described previously.

### Costs of Compliance

We estimate that this proposed AD would affect 53 airplanes of U.S. registry. We also estimate that it would take about 48 work-hours per product to comply with this proposed AD. The average labor rate is \$85 per work-hour. Required parts would cost about \$122,617 per product. Based on these figures, we estimate the cost of this proposed AD to the U.S. operators to be \$6,714,941, or \$122,697 per product.

### Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. "Subtitle VII: Aviation Programs," describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in "Subtitle VII, Part A, Subpart III, Section 44701: General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

### Regulatory Findings

We determined that this proposed AD would not have federalism implications under Executive Order 13132. This proposed AD would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify this proposed regulation:

1. Is not a "significant regulatory action" under Executive Order 12866,
2. Is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979), and
3. Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

You can find our regulatory evaluation and the estimated costs of compliance in the AD Docket.

### List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

### The Proposed Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA proposes to amend 14 CFR part 39 as follows:

### PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

#### § 39.13 [Amended]

2. The FAA amends § 39.13 by adding the following new AD:

**The Boeing Company:** Docket No. FAA–2011–0026; Directorate Identifier 2010–NM–104–AD.

#### Comments Due Date

(a) We must receive comments by March 7, 2011.

#### Affected ADs

(b) None.

#### Applicability

(c) This AD applies to The Boeing Company Model 777–200 and –300 series airplanes, certificated in any category; equipped with Pratt and Whitney engines; as identified in Boeing Alert Service Bulletin 777–54A0024, Revision 1, dated November 4, 2010.

#### Subject

(d) Air Transport Association (ATA) of America Code 54: Nacelles/Pylons.

#### Unsafe Condition

(e) This AD results from reports of system disconnect boxes that have been contaminated with hydraulic fluid, in which one case a crack was found. The Federal Aviation Administration is issuing this AD to detect and correct hydraulic fluid contamination, which can cause cracking of titanium parts in the system disconnect assembly, resulting in compromise of the engine firewall. A cracked firewall can allow fire in the engine area to enter the strut and can lead to an uncontained engine strut fire if flammable fluid is present. Cracking of the disconnect box may also reduce the effectiveness of the fire extinguishing system in the engine compartment and could contribute to an uncontained engine fire. In addition, a cracked disconnect box can leak flammable fluids into the engine core, which can initiate an engine fire and lead to one or both fire conditions discussed above.

#### Compliance

(f) You are responsible for having the actions required by this AD performed within

the compliance times specified, unless the actions have already been done.

### Inspections and Corrective Actions

(g) Within 12 months after the effective date of this AD: Do a general visual inspection for hydraulic fluid contamination of the interior of the strut disconnect assembly, in accordance with Part 1 of the Accomplishment Instructions of Boeing Alert Service Bulletin 777–54A0024, Revision 1, dated November 4, 2010.

(1) For airplanes on which no hydraulic fluid contamination is found (Condition 1): Repeat the general visual inspection required by paragraph (g) of this AD thereafter at intervals not to exceed 6,000 flight cycles or 750 days, whichever occurs first.

(2) For airplanes on which hydraulic fluid contamination is found (Condition 2): Before further flight, do a detailed inspection for discrepancies (e.g., hydraulic fluid coking, heat discoloration, cracks, and etching or pitting) of the interior of the strut disconnect assembly, in accordance with Part 2 of the Accomplishment Instructions of Boeing Alert Service Bulletin 777–54A0024, Revision 1, dated November 4, 2010.

(i) For airplanes on which no discrepancy is found during the inspection required by paragraph (g)(2) of this AD (Condition 2A): Repeat the detailed inspection required by paragraph (g)(2) of this AD thereafter at intervals not to exceed 6,000 flight cycles or 750 days, whichever occurs first.

(ii) For airplanes on which hydraulic fluid coking or heat discoloration is found but no cracking, etching, or pitting is found during the inspection required by paragraph (g)(2) of this AD (Condition 2B): Do the actions required by paragraph (g)(2)(ii)(A) and (g)(2)(ii)(B) of this AD.

(A) Within 300 flight cycles after doing the inspection required by paragraph (g)(2) of this AD: Do a detailed inspection of the exterior of the strut disconnect assembly for cracks, in accordance with Part 3 of the Accomplishment Instructions of Boeing Alert Service Bulletin 777–54A0024, Revision 1, dated November 4, 2010; and repeat the detailed inspection thereafter at intervals not to exceed 300 flight cycles.

(B) Within 6,000 flight cycles or 750 days after hydraulic fluid coking and/or heat discoloration was found during the inspection required by paragraph (g)(2) of this AD, whichever occurs first: Replace the titanium system disconnect assembly with an Inconel system, in accordance with Part 4 of the Accomplishment Instructions of Boeing Alert Service Bulletin 777–54A0024, Revision 1, dated November 4, 2010.

(h) For airplanes on which any crack, etching, or pitting is found during any inspection required by paragraph (g)(2) or (g)(2)(ii)(A) of this AD (Condition 3): Before further flight, replace the titanium system disconnect assembly with an Inconel system, in accordance with Part 4 of the Accomplishment Instructions of Boeing Alert Service Bulletin 777–54A0024, Revision 1, dated November 4, 2010.

### Optional Terminating Action

(i) Replacing the titanium system disconnect assembly with an Inconel system

disconnect assembly in accordance with Part 4 of the Accomplishment Instructions of Boeing Alert Service Bulletin 777-54A0024, Revision 1, dated November 4, 2010, terminates the actions required by this AD.

#### Credit for Actions Accomplished in Accordance With Previous Service Information

(j) Actions accomplished before the effective date of this AD according to Boeing Alert Service Bulletin 777-54A0024, dated April 1, 2010, are considered acceptable for compliance with the corresponding actions specified in this AD.

#### Alternative Methods of Compliance (AMOCs)

(k)(1) The Manager, Seattle Aircraft Certification Office (ACO), FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. Send information to ATTN: Kevin Nguyen, Propulsion Branch, ANM-140S, FAA, Seattle Aircraft Certification Office (ACO), 1601 Lind Avenue, SW., Renton, Washington 98057-3356; telephone (425) 917-6501; fax (425) 917-6590. Information may be e-mailed to: [9-ANM-Seattle-ACO-AMOC-Requests@faa.gov](mailto:9-ANM-Seattle-ACO-AMOC-Requests@faa.gov).

(2) Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office.

Issued in Renton, Washington, on January 12, 2011.

**Jeffrey E. Duven,**

*Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.*

[FR Doc. 2011-1119 Filed 1-19-11; 8:45 am]

**BILLING CODE 4910-13-P**

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 71

[Docket No. FAA-2010-1209; Airspace Docket No. 10-ANM-10]

#### Proposed Amendment of Class E Airspace; West Yellowstone, MT

**AGENCY:** Federal Aviation Administration (FAA), DOT.

**ACTION:** Notice of proposed rulemaking (NPRM).

**SUMMARY:** This action proposes to modify Class E airspace at Yellowstone Airport, West Yellowstone, MT, to accommodate aircraft using the Instrument Landing System (ILS) Localizer (LOC) standard instrument approach procedures at Yellowstone Airport, West Yellowstone, MT. The FAA is proposing this action to enhance the safety and management of aircraft operations at Yellowstone Airport.

**DATES:** Comments must be received on or before March 7, 2011.

**ADDRESSES:** Send comments on this proposal to the U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue, SE., Washington, DC 20590; telephone (202) 366-9826. You must identify FAA Docket No. FAA-2010-1209; Airspace Docket No. 10-ANM-10, at the beginning of your comments. You may also submit comments through the Internet at <http://www.regulations.gov>.

#### FOR FURTHER INFORMATION CONTACT:

Eldon Taylor, Federal Aviation Administration, Operations Support Group, Western Service Center, 1601 Lind Avenue, SW., Renton, WA 98057; telephone (425) 203-4537.

#### SUPPLEMENTARY INFORMATION:

##### Comments Invited

Interested parties are invited to participate in this proposed rulemaking by submitting such written data, views, or arguments, as they may desire. Comments that provide the factual basis supporting the views and suggestions presented are particularly helpful in developing reasoned regulatory decisions on the proposal. Comments are specifically invited on the overall regulatory, aeronautical, economic, environmental, and energy-related aspects of the proposal.

Communications should identify both docket numbers (FAA Docket No. FAA-2010-1209 and Airspace Docket No. 10-ANM-10) and be submitted in triplicate to the Docket Management System (see **ADDRESSES** section for address and phone number). You may also submit comments through the Internet at <http://www.regulations.gov>.

Commenters wishing the FAA to acknowledge receipt of their comments on this action must submit with those comments a self-addressed stamped postcard on which the following statement is made: "Comments to FAA Docket No. FAA-2010-1209 and Airspace Docket No. 10-ANM-10". The postcard will be date/time stamped and returned to the commenter.

All communications received on or before the specified closing date for comments will be considered before taking action on the proposed rule. The proposal contained in this action may be changed in light of comments received. All comments submitted will be available for examination in the public docket both before and after the closing date for comments. A report summarizing each substantive public contact with FAA personnel concerned with this rulemaking will be filed in the docket.

#### Availability of NPRMs

An electronic copy of this document may be downloaded through the Internet at <http://www.regulations.gov>. Recently published rulemaking documents can also be accessed through the FAA's web page at [http://www.faa.gov/airports\\_airtraffic/air\\_traffic/publications/airspace\\_amendments/](http://www.faa.gov/airports_airtraffic/air_traffic/publications/airspace_amendments/).

You may review the public docket containing the proposal, any comments received, and any final disposition in person in the Dockets Office (see the **ADDRESSES** section for the address and phone number) between 9 a.m. and 5 p.m., Monday through Friday, except federal holidays. An informal docket may also be examined during normal business hours at the Northwest Mountain Regional Office of the Federal Aviation Administration, Air Traffic Organization, Western Service Center, Operations Support Group, 1601 Lind Avenue, SW., Renton, WA 98057.

Persons interested in being placed on a mailing list for future NPRMs should contact the FAA's Office of Rulemaking, (202) 267-9677, for a copy of Advisory Circular No. 11-2A, Notice of Proposed Rulemaking Distribution System, which describes the application procedure.

#### The Proposal

The FAA is proposing an amendment to Title 14 Code of Federal Regulations (14 CFR) Part 71 by modifying Class E airspace extending upward from 700 feet above the surface at Yellowstone Airport, West Yellowstone, MT, to accommodate new ILS LOC standard instrument approach procedures at Yellowstone Airport, West Yellowstone, MT. This action would enhance the safety and management of aircraft operations at Yellowstone Airport.

Class E airspace designations are published in paragraph 6005, of FAA Order 7400.9U, dated August 18, 2010, and effective September 15, 2010, which is incorporated by reference in 14 CFR 71.1. The Class E airspace designation listed in this document will be published subsequently in this Order.

The FAA has determined this proposed regulation only involves an established body of technical regulations for which frequent and routine amendments are necessary to keep them operationally current. Therefore, this proposed regulation; (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034; February 26, 1979); and (3) does not warrant preparation of a regulatory evaluation as the anticipated